

Abstract:

The invention relates to a gas turbine vane, especially a vane pertaining to an aircraft engine, comprising a blade (11) and a vane footing (12). Said blade (11) is defined by a flow inlet edge or a front edge (13), a flow outlet edge or a rear edge (14), and a blade surface (15) extending between the front edge (13) and the rear edge (14) and forming a suction side (16) and a pressure side (17). According to the invention, the suction side (17) of the blade (11) comprises at least one microprofiled or microstructured region (18; 20, 21, 22) for optimizing the flow around the blade (11).